**Changes to make to successorPC work starting from the AeroDyn segmentation fault error.**

*Error log:* 

1. Change the file AeroDyn\_blade.dat and delete the last row. Notice that it implies a very small displacement along the span and a significant increase in chord.

A screen shot of a computer

Description automatically generated

1. Change the AeroDyn15.dat file in the Airfoil information section. Add the variable AFTabMod and set it =2.

A black background with many squares

Description automatically generated with medium confidence

1. Make sure you are copying the boundaryData folder and not the boundaryDataPre folder into the successor’s constant folder. If not, it won’t be able to read the inlet values.

A black background with white text

Description automatically generated CORRECT

A black background with white text

Description automatically generated INCORRECT

1. This case (successorPC) HAS NO REFINEMENT. No refinemeshLocal, no snappyHexMesh, no mapFields, no renumberMesh. Only blockmesh and changeDictionary
2. Initializer (setFieldsABL) has to be run before running the solver
3. PISO has to be substituted by PIMPLE in system/fvSolution
4. In system/fvSchemes, the sub-dictionary fluxRequired has to be changed. Instead of pressure (p), substitute by (p\_rgh)\*
5. Remove all sampling functions from controlDict.1 because they make reference to sampling points which do not correspond with the domain in successorPC.